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Design Process Formulas

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List of 19 Design Process Formulas

Design Process

1) Battery Weight Fraction

$$fx \quad WBF = \left(\frac{R}{E_{\text{battery}} \cdot 3600 \cdot \eta \cdot \left(\frac{1}{[g]} \right) \cdot LD_{\text{maxratio}}} \right)$$

[Open Calculator !\[\]\(a870788d6ed9b8fd294b7654a8c8526b_img.jpg\)](#)

$$ex \quad 0.054049 = \left(\frac{10\text{km}}{21\text{J/kg} \cdot 3600 \cdot 0.80 \cdot \left(\frac{1}{[g]} \right) \cdot 30} \right)$$

2) Cost Index given Minimum Design Index

$$fx \quad CI = \frac{(DI_{\text{min}} \cdot 100) - (WI \cdot P_w) - (TI \cdot P_t)}{P_c}$$

[Open Calculator !\[\]\(c50c8b7b2cc2cf9ff925edec0ee94c0d_img.jpg\)](#)

$$ex \quad 1327.913 = \frac{(160 \cdot 100) - (50.98 \cdot 15.1) - (95 \cdot 19)}{10.11}$$

3) Electric Power for Wind Turbine

$$fx \quad P_e = W_{\text{shaft}} \cdot \eta_g \cdot \eta_{\text{transmission}}$$

[Open Calculator !\[\]\(f60b7a900783ac3fd531bfd9c111be6d_img.jpg\)](#)

$$ex \quad 0.192\text{kW} = 0.6\text{kW} \cdot 0.8 \cdot .4$$

4) Fuel Load

$$fx \quad W_f = W_{\text{misf}} + W_{\text{resf}}$$

[Open Calculator !\[\]\(83bbbd261710c59db0214aa27b2edc0d_img.jpg\)](#)

$$ex \quad 9499\text{kg} = 8761\text{kg} + 738\text{kg}$$




5) Induced Inflow Ratio in Hover 

$$fx \quad \lambda = \frac{V_i}{R_{rotor} \cdot \omega}$$

Open Calculator 

$$ex \quad 4.142857 = \frac{58\text{m/s}}{0.007\text{km} \cdot 2\text{rad/s}}$$

6) Maximum Payload Capability 

$$fx \quad W_{pay} = MTOW - W_{OE} - W_f$$

Open Calculator 


$$ex \quad 52370\text{kg} = 62322\text{kg} - 453\text{kg} - 9499\text{kg}$$

7) Minimum Design Index 

$$fx \quad DI_{min} = \frac{(CI \cdot P_c) + (WI \cdot P_w) + (TI \cdot P_t)}{100}$$

Open Calculator 

$$ex \quad 160 = \frac{(1327.913 \cdot 10.11) + (50.98 \cdot 15.1) + (95 \cdot 19)}{100}$$

8) Mission Fuel 

$$fx \quad W_{misf} = W_f - W_{resf}$$

Open Calculator 

$$ex \quad 8761\text{kg} = 9499\text{kg} - 738\text{kg}$$

9) Period of Design Index given Minimum Design Index 

$$fx \quad TI = \frac{(DI_{min} \cdot 100) - (WI \cdot P_w) - (CI \cdot P_c)}{P_t}$$

Open Calculator 

$$ex \quad 95.00008 = \frac{(160 \cdot 100) - (50.98 \cdot 15.1) - (1327.913 \cdot 10.11)}{19}$$



10) Priority of Objective Cost in Design Process given Minimum Design Index 

$$fx \quad P_c = \frac{(DI_{\min} \cdot 100) - (WI \cdot P_w) - (TI \cdot P_t)}{CI}$$

[Open Calculator !\[\]\(e78f798d4ea5c530c9db49e7d26e6b95_img.jpg\)](#)

$$ex \quad 10.11 = \frac{(160 \cdot 100) - (50.98 \cdot 15.1) - (95 \cdot 19)}{1327.913}$$

11) Priority of Objective Period of Design given Minimum Design Index 

$$fx \quad P_t = \frac{(DI_{\min} \cdot 100) - (WI \cdot P_w) - (CI \cdot P_c)}{TI}$$

[Open Calculator !\[\]\(05be7c7a8995decd503647c99211f7c2_img.jpg\)](#)

$$ex \quad 19.00002 = \frac{(160 \cdot 100) - (50.98 \cdot 15.1) - (1327.913 \cdot 10.11)}{95}$$

12) Priority of Objective Weight in Design Process given Minimum Design Index 

$$fx \quad P_w = \frac{(DI_{\min} \cdot 100) - (CI \cdot P_c) - (TI \cdot P_t)}{WI}$$

[Open Calculator !\[\]\(fe3aebe81acea8d45108cd2768939da7_img.jpg\)](#)

$$ex \quad 15.10003 = \frac{(160 \cdot 100) - (1327.913 \cdot 10.11) - (95 \cdot 19)}{50.98}$$

13) Propulsion Net Thrust 

$$fx \quad Ft = m_{af} \cdot (V_J - V_f)$$

[Open Calculator !\[\]\(899d8b7697d64725bf017d3296cfcf1b_img.jpg\)](#)

$$ex \quad 9.81N = 0.9kg/s \cdot (60.90m/s - 50m/s)$$

14) Range Increment of Aircraft 

$$fx \quad \Delta R = R_D - R_H$$

[Open Calculator !\[\]\(40770d9ed6ed4f1222ebf89a1396e8b2_img.jpg\)](#)

$$ex \quad 334km = 1220km - 886km$$



15) Reserve Fuel 

$$fx \quad W_{\text{resf}} = W_f - W_{\text{misf}}$$

[Open Calculator !\[\]\(e2376d476d06eb31946dc01a69a4403a_img.jpg\)](#)


$$ex \quad 738\text{kg} = 9499\text{kg} - 8761\text{kg}$$

16) Summation of Priorities of all Objectives that need to be Minimized 

$$fx \quad P_{\text{min}} = P_c + P_w + P_t$$

[Open Calculator !\[\]\(0b5e7e25e8775f7e7e80906ada4f0021_img.jpg\)](#)

$$ex \quad 44.21 = 10.11 + 15.1 + 19$$

17) Summations of Priorities of Objectives that need to be Maximized (Military planes) 

$$fx \quad P_{\text{max}} = P_p + P_f + P_b + P_m + P_r + P_d + P_s$$

[Open Calculator !\[\]\(bd3b31712ad9bab5a241210fa6925cdd_img.jpg\)](#)

$$ex \quad 76 = 11 + 14 + 10.5 + 6 + 13 + 12 + 9.5$$

18) Thrust-to-Weight Ratio given Vertical Velocity 

fx

[Open Calculator !\[\]\(7bc43b319a082987e20f7bf78f4bab80_img.jpg\)](#)

$$TW = \left(\left(\frac{V_v}{V_a} \right) + \left(\left(\frac{P_{\text{dynamic}}}{W_S} \right) \cdot (C_{D\text{min}}) \right) + \left(\left(\frac{k}{P_{\text{dynamic}}} \right) \cdot (W_S) \right) \right)$$

$$ex \quad 17.96714 = \left(\left(\frac{54\text{m/s}}{206\text{m/s}} \right) + \left(\left(\frac{8\text{Pa}}{5\text{Pa}} \right) \cdot (1.3) \right) + \left(\left(\frac{25}{8\text{Pa}} \right) \cdot (5\text{Pa}) \right) \right)$$

19) Weight Index given Minimum Design Index 

$$fx \quad WI = \frac{(DI_{\text{min}} \cdot 100) - (CI \cdot P_c) - (TI \cdot P_t)}{P_w}$$

[Open Calculator !\[\]\(e119fc79c8f448683d20ba4c873025a2_img.jpg\)](#)

$$ex \quad 50.9801 = \frac{(160 \cdot 100) - (1327.913 \cdot 10.11) - (95 \cdot 19)}{15.1}$$



Variables Used










- C_{Dmin} Minimum Drag Coefficient
- CI Cost Index
- DI_{min} Minimum Design Index
- $E_{battery}$ Battery Specific Energy Capacity (*Joule per Kilogram*)
- F_t Thrust Force (*Newton*)
- k Lift Induced Drag Constant
- $LD_{max, ratio}$ Maximum Lift to Drag Ratio of Aircraft
- m_{af} Air Mass Flow Rate (*Kilogram per Second*)
- $MTOW$ Maximum Take Off Weight (*Kilogram*)
- P_b Scariness Priority (%)
- P_c Cost Priority (%)
- P_d Disposability Priority (%)
- $P_{dynamic}$ Dynamic Pressure (*Pascal*)
- P_e Electric Power of Wind Turbine (*Kilowatt*)
- P_f Flight Quality Priority (%)
- P_m Maintainability Priority (%)
- P_{max} Priority Sum of Objectives to be Maximized (%)
- P_{min} Priority Sum of Objectives to be Minimized(%)
- P_p Performance Priority (%)
- P_r Producibility Priority (%)
- P_s Stealth Priority (%)
- P_t Period Priority (%)
- P_w Weight Priority (%)
- R Range of Aircraft (*Kilometer*)
- R_D Design Range (*Kilometer*)
- R_H Harmonic Range (*Kilometer*)



- **R_{rotor}** Rotor Radius (Kilometer)
- **TI** Period Index
- **TW** Thrust-to-Weight Ratio
- **V_a** Aircraft Velocity (Meter per Second)
- **V_f** Flight Velocity (Meter per Second)
- **v_i** Induced Velocity (Meter per Second)
- **V_J** Velocity of Jet (Meter per Second)
- **V_v** Vertical Airspeed (Meter per Second)
- **W_f** Fuel Load (Kilogram)
- **W_{misf}** Mission Fuel (Kilogram)
- **W_{OE}** Operating Empty Weight (Kilogram)
- **W_{pay}** Payload (Kilogram)
- **W_{resf}** Reserve Fuel (Kilogram)
- **W_S** Wing Loading (Pascal)
- **W_{shaft}** Shaft Power (Kilowatt)
- **WBF** Battery Weight Fraction
- **WI** Weight Index
- **ΔR** Range Increment of Aircraft (Kilometer)
- **η** Efficiency
- **η_g** Efficiency of Generator
- **$\eta_{\text{transmission}}$** Efficiency of Transmission
- **λ** Inflow Ratio
- **ω** Angular Velocity (Radian per Second)



Constants, Functions, Measurements used

- **Constant:** [g], 9.80665
Gravitational acceleration on Earth
- **Measurement: Length** in Kilometer (km)
Length Unit Conversion 
- **Measurement: Weight** in Kilogram (kg)
Weight Unit Conversion 
- **Measurement: Pressure** in Pascal (Pa)
Pressure Unit Conversion 
- **Measurement: Speed** in Meter per Second (m/s)
Speed Unit Conversion 
- **Measurement: Power** in Kilowatt (kW)
Power Unit Conversion 
- **Measurement: Force** in Newton (N)
Force Unit Conversion 
- **Measurement: Mass Flow Rate** in Kilogram per Second (kg/s)
Mass Flow Rate Unit Conversion 
- **Measurement: Angular Velocity** in Radian per Second (rad/s)
Angular Velocity Unit Conversion 
- **Measurement: Specific Energy** in Joule per Kilogram (J/kg)
Specific Energy Unit Conversion 



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